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Solution-1-H6739.tex 24/1/2007 9:28 Page6. 6 Solutions Manual Fig. S.1.3(c) Fig. S.1.3(d) S.1.4. The principal stresses at the point are determined, as indicated in the question, by transforming each state of stress into a $\sigma_x, \sigma_y, \tau_{xy}$ stress system. Clearly, in the first case $\sigma_x=0, \sigma_y=10\text{N/mm}^2, \tau_{xy}=0$ (Fig. S.1.4(a)).

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Aircraft Structures for Engineering Students, Fourth Edition
Aircraft Structures for Engineering Students, Sixth Edition. The major modification in the sixth edition is the extension of the work on composite materials and structures presented in Chapter 25. In the fifth edition, the theory was restricted to single-ply laminates; this has now been extended to a consideration of multi-ply laminates.

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